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GROUP 3400

IN THE UNITED STATES PATENT AND TRADEMARK OFFICE

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Applicants: Mannava, et al.)
Serial No: 08/719,341) Group Art Unit: 3745
Filed: September 25, 1996) Examiner: C. Verdier
For: LASER SHOCK PEENED GAS TURBINE)
ENGINE COMPRESSOR AIRFOIL EDGES)

Hon. Assistant Commissioner for Patents
Washington, D.C. 20231

SIR:

PRELIMINARY AMENDMENT

In accordance with your request made during a telephone conversation with attorney Steven J. Rosen on February 2, 1999, please enter the Declaration filed under 37 C.F.R. 1.131(b) and dated October 7, 1998 and please consider and enter the following Amendments.

IN THE CLAIMS:

Please amend the following claims as indicated.

1. (FIVE TIMES AMENDED) A gas turbine engine component comprising:

a metallic compressor airfoil having a leading edge and a trailing edge and a pressure side and a suction side,

at least a first laser shock peened surface on a first side of said airfoil,

said laser shock peened surface extending radially along at least a portion of said leading edge and extending chordwise from said leading edge, and

a first region having deep compressive residual stresses imparted by laser shock peening (LSP) extending into said airfoil from said laser shock peened surface wherein said deep

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